

The 6-meter Breadboard Antenna for the Deep Space Network Large Array¹

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Abstract—Development of very large arrays of small antennas has been proposed as a way to increase the downlink capability of the NASA Deep Space Network (DSN) by two or three orders of magnitude thereby enabling greatly increased science data from currently configured missions or enabling new mission concepts. The current concept is for an array of 400 x 12-m antennas at each of three longitudes [1]. The DSN array will utilize radio astronomy sources for phase calibration and will have wide bandwidth correlation processing for this purpose. A program is currently underway to develop the technology and prove the performance and cost of a very large DSN array. The program includes a 3-element interferometer to be completed by late 2004. This paper describes the design and development of the low cost 6-meter breadboard antenna to be used as part of the interferometer.

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1. INTRODUCTION

The baseline breadboard antenna is a *6-meter hydroformed symmetrically shaped dual reflector system utilizing Gregorian optics*. The heritage for this antenna lies in the Allen Telescope Array project plans to build 350 6-m antennas specified for use up to 11 GHz. Hydroforming is the process of forming aluminum to a rigid and precise mold by using a fluid or gas under pressure. It has been highly developed for use in production of low-cost reflectors for satellite communications and thousands of antennas in the 1 to 4 meter range have been manufactured. This project improves the rms of the surface to extend the useable

frequency range to 40 GHz. The backup structure utilizes 9 equally spaced aluminum struts connecting a center yoke to the rim of the dish. The pedestal consists of a central pipe tucked under the dish with a central bearing for azimuth motion and a jackscrew for elevation control.

Also under consideration, as an alternative design for the hydroformed dishes, is an antenna manufactured from composite materials. Composite materials have proven mechanical and structural properties that are superior to metals, at a reduced weight. The major challenge with this type of structure is the cost. But new manufacturing processes and novel implementations are expected to reduce the cost of manufacturing considerably. Other challenges that need to be resolved include the environmental performance and finding a suitable material to incorporate as the reflective surface.

The feed is a dual frequency horn covering the X (8–9 GHz) and Ka (30–40 GHz) bands. The X-band will be a coaxial waveguide fed corrugated horn and Ka-band will utilize a dielectric rod in the center of the corrugated horn.

2. RF OPTICS DESIGN

An early design for the antenna proposed by Vahraz Jamnejad [2] was designed for optimal gain, not the desired optimal gain to noise temperature ratio. His design demonstrated a gain of 52.82 dB and a G/T of 38.49 dB/K (assuming a 240 K ambient temperature and an amplifier noise temperature of 15 K). It was calculated that an antenna designed for optimal G/T, however, would yield a G/T of 40.9 dB/K. Unfortunately, the main reflector profile of Jamnejad's antenna had already been submitted for fabrication. Therefore, the antenna was redesigned for the maximum G/T attainable, by reshaping the subreflector and moving the feed location, while retaining the main reflector design.

Optimizing for G/T

In a dual reflector antenna geometrical optics shaped for maximum gain, the main reflector is illuminated by the subreflector in such a way as to produce a uniform aperture distribution [3]. This utilizes a subreflector pattern that has a high edge taper that is truncated to zero at the edge of the

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main reflector. Unfortunately, due to diffraction effects, a real subreflector pattern does not go to zero at the main reflector edge and there is substantial spillover in the rear direction. This spillover sees the hot earth and consequently increases the noise temperature of the antenna system. The DSN has typically dealt with this problem in two ways. 1) Select the uniform illumination function of the main reflector to be less than the physical aperture, thus using the remainder of the aperture as a noise shield thus reducing the spillover energy that falls on the hot earth or 2) Select the illumination function to be uniform to a give radius and then taper it to zero at the reflector edge, also reducing the rear spillover. The 70-meter antennas, the HEF, DSS-13 and the ARST antennas used method 1) and the operational BWG antennas used method 2). Both methods yield virtually identical results for G/T.

Since the problem with the maximum gain design is the rear spillover, it is possible to reduce the rear spillover by truncating the subreflector and letting the power spill in the forward direction, which only sees the cold sky. However, the reduction in subreflector size necessary to significantly reduce the real spillover loses a significant fraction of the gain to forward spillover. The maximum G/T obtainable using this method is about 1 dB below the optimum G/T.

Another method to reduce the forward spillover is to move the feed forward of the design position and then reshape the subreflector for perfect phase in the aperture. Selecting the amount of the main reflector illuminated by the subreflector controls the rear spillover. The position of the feed and the amount of the main reflector illuminated are the two parameters to be optimized. For each feed location and illumination radius, the subreflector was reshaped for uniform phase in the aperture. Figure 1 summarizes the results of the simulations.

Not included in the analysis was the feed blockage from the subreflector-scattered field. In order to ensure against undesired blockage from the horn with a displacement too extreme, a 0.2 m maximum displacement was chosen. The corresponding main reflector uniform illumination that yielded the greatest G/T, a 2.9 m illumination radius, was then chosen. This design demonstrates a gain of 53.56 dB and a G/T of 40.77 dB/K at 8.4 GHz, a 2.28 dB/K G/T increase from the original design.

The above calculations were based upon a feed phase center selected for maximum gain. Optimizing the feed phase center location for maximum G/T then further increased this performance. After performing analysis to determine the optimal feed position for the antenna, we determined that a 0.00508 m offset yielded optimal performance over the bandwidth. This further increased the G/T to 40.82 dB/K, with an associated gain of 53.50 dB. This final design yields a 2.33 dB/K increase of G/T from the original design.

Figure 2 demonstrates the performance of the final antenna; which has 0.2 m feed displacement for the original design, a

2.9 m uniform illumination radius, and 0.00508 m feed phase center offset. The chart has the data with the calculated radiation patterns of the feed as well as some measured feed patterns in the 89 GHz and 3040 GHz bandwidths.

G/T Estimates

The above calculations were done primarily for tradeoff comparisons and did not include all the estimated losses that would be common to all designs. The above results included the calculated losses from the PO programs and an estimated 15K noise temperature contribution from the low noise amplifier system at X-band and 40K at Ka-band. The purpose of this section is to provide a more complete G/T performance estimate including the expected uncertainties.

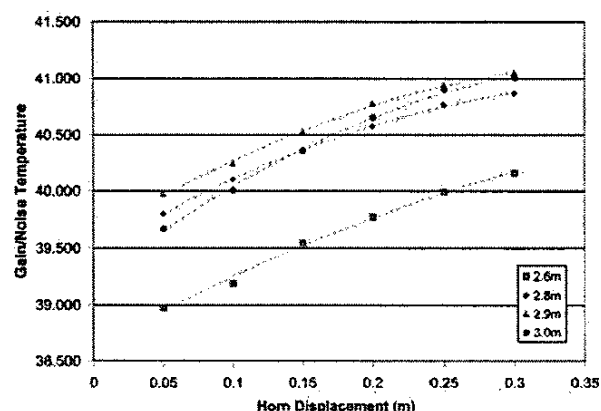


Figure 1 - G/T vs. Horn Displacement for Various Illumination Radius

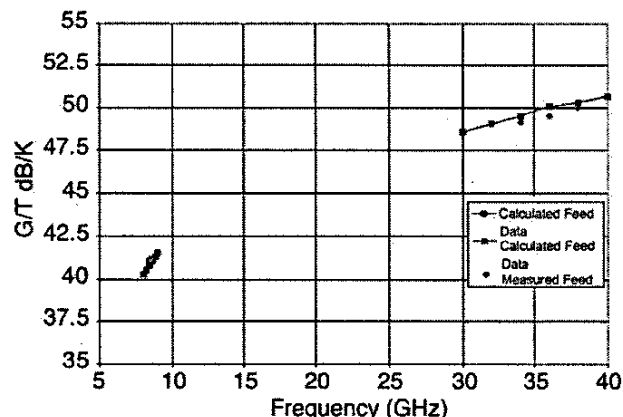


Figure 2 - G/T Performance Across Bandwidth

Tables 1 and 2 give the performance estimates for the X- and Ka-band low noise amplifiers. These are the wideband MMIC design. A typical estimate for the system noise temperature is shown in Table 3 and a typical gain budget shown in Table 4. Utilizing the data from these tables along with the PO calculated gain as a function of frequency utilizing the theoretical feed patterns, the maximum and minimum estimate G/T is shown in Figure 3. Further details on sensitivity performance can be found in Reference [4].

Table 1. X-band Noise Budget (S. Petty/M. Britcliffe)

Item	Loss L (dB)		Phy Temp T, K		Gain G, dB		Tin, K		Noise, K	
	Goal	Max	Goal	Max	Goal	Max	Goal	Max	Goal	Max
Mismatch loss, vacuum window (included in vacuum window resistive loss term below)									0.00	0.00
X-band combiners isolation noise term. Isolation = 26 dB goal, 23 dB max									0.03	0.06
X-band polarizer isolation noise term. Isolation = 23 dB goal, 20 dB max									0.06	0.12
Vacuum window	0.004	0.015	293	293					0.29	1.00
Goal: 11 mil Teflon, 3 mil Kapton, 1" Propozote										
Max: 11 mil Teflon, 0.25" Rexolite, 1" Propozote										
Feedhorn	0.008	0.014	50	60					0.10	0.19
Teflon torpedo/foam support	0.015	0.020	20	30					0.07	0.14
Waveguide Cu 1.5 inches	0.002	0.003	15.0	16.0					0.08	0.19
Slot combiner	0.025	0.050	13.0	15.0					0.08	0.18
WR112 7.5 inches straight, 2 miter bends	0.027	0.050	13.0	15.0					0.08	0.18
Slot combiner	0.025	0.050	13.0	15.0					0.08	0.18
Hybrid polarizer	0.030	0.050	13.0	15.0					0.09	0.18
WR bend	0.006	0.008	13.0	14.0					0.02	0.03
WR cal coupler (loss)	0.025	0.035	13.0	14.0					0.08	0.12
WR cal coupler (inj noise)	30.0	29.0	293	293					0.30	0.39
Goal: Coupler integrated in MMIC module										
Max: Coupler separate (same performance)										
WR/SMA male adapter	0.090	0.120	13.0	6.9					0.28	0.21
MMIC HEMT module					35.0	31.0	4.00	7.00	4.24	7.70
0.141 output coax, 12-70K	0.56	0.67	50	50					0.00	0.01
0.141 output coax, 70-293K	0.56	0.67	210	210					0.01	0.04
Loss between vacuum feed through, receiver assembly	0.40	0.50	293	293					0.01	0.04
Receiver assembly							150	250	0.15	0.90
Input Noise Temp Te(K)									6.14	12.35

Table 2. Ka-band Noise Budget (S. Petty/M. Britcliffe)

Item	Loss L (dB)		Phy Temp T, K		Gain G, dB		Tin, K		Noise, K	
	Goal	Max	Goal	Max	Goal	Max	Goal	Max	Goal	Max
Mismatch loss, vacuum window (included in vacuum window resistive loss term below)									0.00	0.00
Ka-band polarizer isolation noise term. Isolation = 20 dB goal, 17 dB max									0.25	0.35
Vacuum window	0.015	0.038	293	293					1.01	2.57
Goal: 11 mil Teflon, 3 mil Kapton, 1" Propozote										
Max: 11 mil Teflon, 0.25" Rexolite, 1" Polystyrene										
Feedhorn	0.020	0.032	50	60					0.23	0.45
Teflon torpedo/foam support	0.045	0.060	20	30					0.21	0.42
Round WG Cu 4.0 inches	0.004	0.006	15.0	18.0					0.01	0.03
Hybrid polarizer	0.050	0.075	13.0	15.0					0.15	0.27
WR28 waveguide bend	0.005	0.005	13.0	15.0					0.02	0.02
WR28 30 dB cal coupler (loss)	0.060	0.060	13.0	15.0					0.19	0.22
WR28 30 dB cal coupler (inj noise)	30.0	29.0	293	293					0.31	0.39
Goal: Coupler integrated in MMIC module										
Max: Coupler separate										
MMIC HEMT module					40.0	38.0	15	22	15.70	23.44
Output coax, 12-70K, 8 in. long	2.00	2.50	50	50					0.00	0.01
Output coax, 70-293K, 8 in. long	2.00	2.50	210	210					0.02	0.05
Loss between vacuum feed through, receiver assembly	1.00	1.40	293	293					0.02	0.06
Receiver assembly							600	800	0.42	1.59
Input Noise Temp Te(K)									18.6	30.2

Table 3. Typical Noise Temperature Budget

Element	Noise, K		Note
	X-band, 8.4 GHz	K-band, 32 GHz	
Cosmic background	2.5	2.0	Effective blackbody
Atmosphere	2.2	7.0	Goldstone (ave clear)
Forward spill	0.3	0	6% at X-band
Main reflector rear spill	3.6	1.0	
Main reflector I ² R	0.1	0.2	Aluminum
Subreflector I ² R	0.1	0.2	Aluminum
Quadripod scatter	2/4	2/4	Estimated
Feed/amplifier cont	6.1/12.4	18.6/30.2	Tables 1 and 2
Total Noise, K	16.9/25.2	31.0/44.6	

Table 4. Typical Efficiency Budget

Element	Efficiency		Note
	X-band, 8.4 GHz	K-band, 32 GHz	
P.O. computed	0.777	0.780	100% = 54.59 -X 100% = 66.21 -Ka
Main Reflector I ² R	0.999	0.999	12 mils RMS
RMS	0.988	0.846	
Subreflector I ² R	0.999	0.999	4 mils RMS
RMS	0.999	0.8982	
Feed support blockage	0.85/0.9	0.85/0.9	Estimated
Feed VSWR	0.999	0.999	
Efficiency	0.650/0.688	0.549/0.581	
Gain (dB)	52.72/52.97	63.61/63.85	

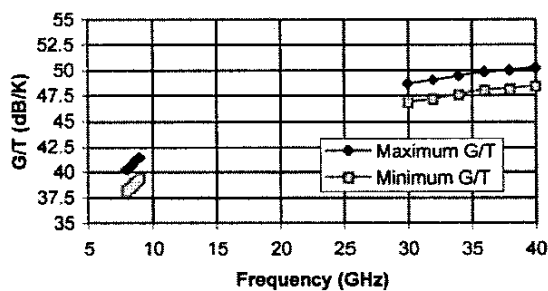


Figure 3 - Maximum and Minimum G/T

3. FEED DESIGN

The basic requirements for the feed are to optimally cover the DSN X- (8.4–8.5 GHz) and Ka- (31.8–32.2 GHz) frequency receive bands and provide useable performance over the 8–9 GHz and 30–40 GHz frequency bands. The polarization is dual circular at both bands with ellipticity

<0.75 dB over the DSN bands. The return loss is to be better than 20 dB over the useable bands. The target illumination function of –12 dB at 42 degrees off main beam direction was selected to be the same as the ATA feed so there was the possibility of using the ATA offset reflectors. Considerations for low cost mass production were also to be included in the design.

After exploring several options, the selected design was an X/Ka band coaxial feed that includes a dielectric rod for Ka-band radiation as shown on Figure 4. The radiation pattern at X-band was controlled by slot depth and flare angle to produce saturated operation and the radiation pattern at Ka-band controlled by the dielectric rod profile. X-band enters the horn throat via a TE₁₁ coax mode and Ka-band enters conventionally. Both bands use a commercial fin type polarizer. A picture of a linear polarization prototype feed is shown in Figure 5. More details of the feed design and performance can be found in Reference [5].

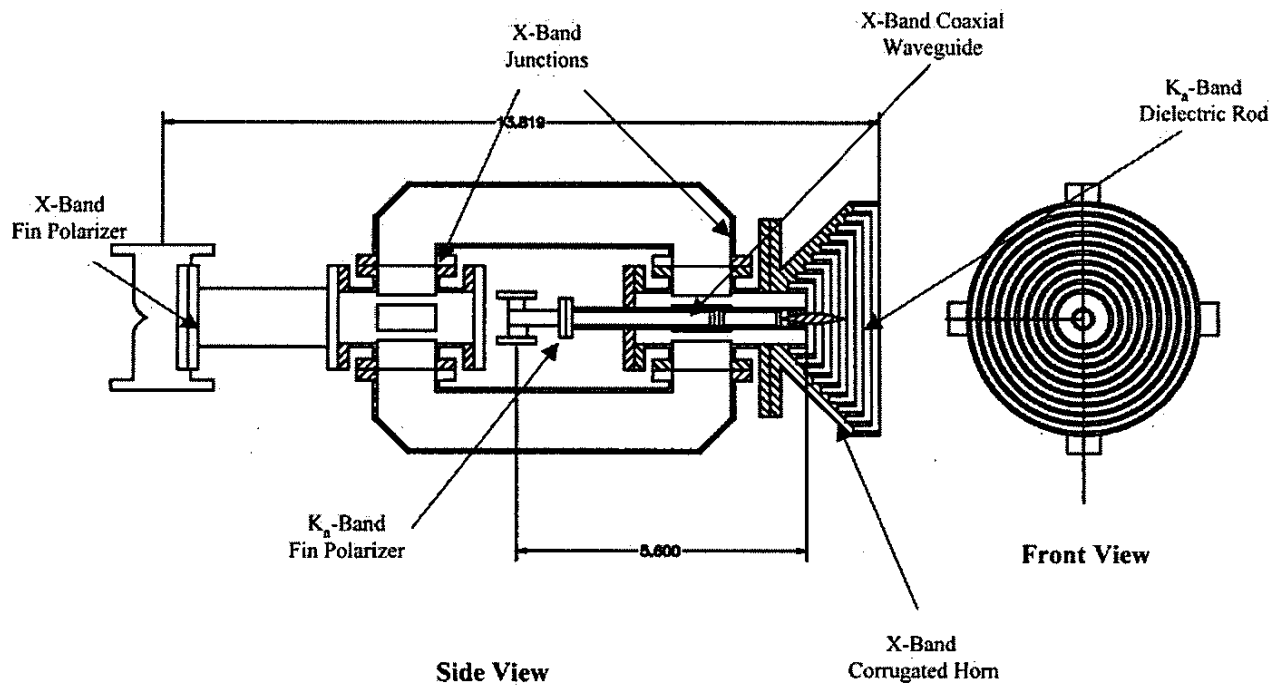


Figure 4 - The X/Ka Band Coaxial Feed

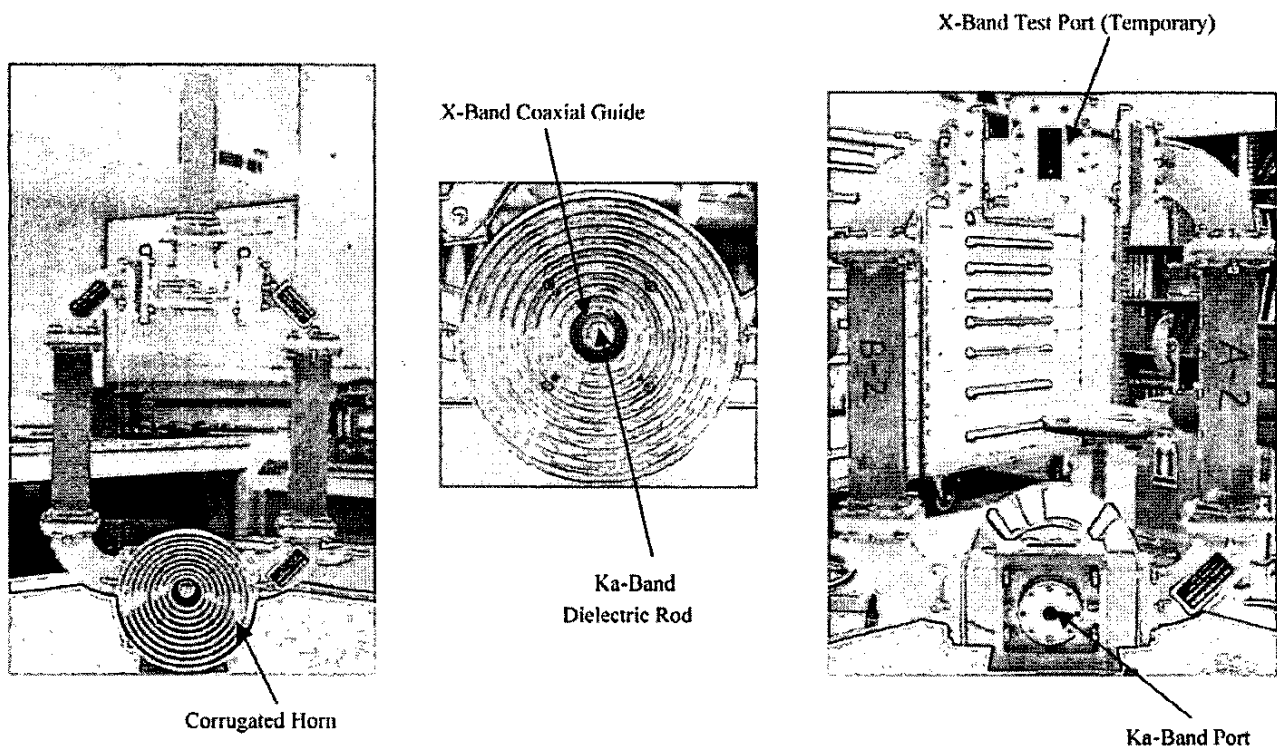


Figure 5 - Prototype Feed – Linear Polarization Test Model

4. CRYOGENIC DESIGN

The following describes the design and performance of a prototype LNA system for The Deep Space Network (DSN) Large Array Task. The system cools a dual frequency feed system equipped with HEMT low noise amplifiers and the associated support electronics. The system was designed to be manufactured at minimum cost. The design considerations including the cryocooler, the vacuum system, microwave interconnects, mechanical components and radiation shielding are discussed. Further details can be found in reference [6].

Cryocooler Selection

One challenge in the design of the large array LNA system is minimizing the cost of electrical power. Estimates of the operations cost of an array with 100's to 1000's of array elements show that power costs for LNA cooling are the biggest single operating cost of the LNA system. Two refrigerators showed promise. Both were Gifford-McMahon two stage coolers with first stage operating temperatures near 50K and second stage temperatures near 15K.

The CTI Cryogenics model 350 are systems that produced 3 watts of cooling at 15K. The input power requirement is 1.8 kW. The design has been used in hundreds of LNA systems over a 40-year span. The units typically operate for 18000 hours before requiring maintenance.

The second was Cryomech Inc. model GB-15 systems. The GB-15 is similar to the original Gifford-McMahon cooler evaluated by JPL in the 1960's. These are Gifford-McMahon (GM) systems that produce 1.5 watts of cooling at 15K. The power consumption is 1.2 kW.

Cryogenic Heat Load

The heat load on the cryocooler is the sum of load imposed by the LNA and supporting components from any residual gas in the vacuum space. Conduction load is through solid supports, wires, signal cables and the residual gas in the vacuum space. Radiation is from the room temperature surroundings to the cryogenic components.

The first stage of the cooler intercepts the radiation load from room temperature to the cooled components with the radiation shield. The first stage is also used to minimize heat conducted to the second stage by thermally connecting solids connected between room temperature and the second stage.

This system includes a cryogenically cooled feed and feed horn. The horn has a 12 cm aperture that is subject to radiation from ambient. The 12 cm feed is the largest cooled aperture that has been used in the DSN in a closed-cycle cooling system.

The system imposes a heat load of 10 watts on the first stage and 1 watt on the second stage of the cooler. Radiation

loading from the cooled feed accounts for over 50 percent of the 2nd stage heat load.

Mechanical Layout

To evaluate both coolers the prototype cooler was designed to accept both the CTI 350 and the Cryomech GB-15. The package is housed in 25 cm diameter 42cm long round vacuum housing. The feed and cooler are offset to provide the most usable space in the cold volume. The assembly is designed to use the Cryomech GB-15 and CTI model 350 coolers interchangeably with only bolt-in modification.

The vacuum housing chosen for the breadboard system is constructed using aluminum tubing. It eliminates welding and the problems associated with vacuum leaks and warping. In production runs this technique can be cost effective. It provides the benefit of minimum weight and allows the vacuum housing to be sized optimally.

The end plate for the vacuum housing uses the standard machined aluminum plate design. The plate is fitted with simple gland-seal O-ring connections. It uses commercial coax interface connectors. The feed is supported mechanically from the base plate with G-10 fiberglass supports. The feed installed is shown in Figure 6. There is no mechanical connection from the opposite end plate. This design has several advantages: It provides minimal heat conduction to cooled components. It allows the package to be easily assembled and minimizes mechanical stress due to thermal expansion.

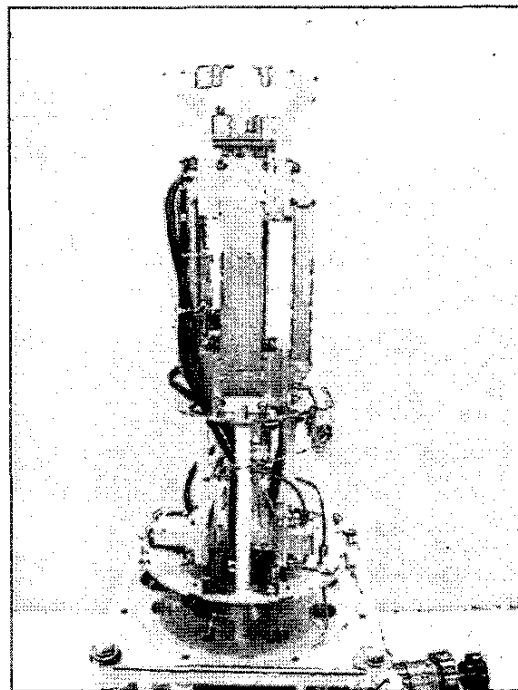


Figure 6 - Feed Installed in LNA Assembly

In an effort to reduce manufacturing costs the mechanical hardware was constructed using sheet metal technology wherever possible. The radiation shield is a simple rolled cone. It is attached with screws to the flanges that are also simple sheet metal parts. It eliminates costly and time consuming welding or silver brazing used in traditional DSN designs. Normally copper is used for shields. The radiation shields for the prototype are constructed of 1100F alloy aluminum. The aluminum provides acceptable thermal conductivity for the application and provides higher resistance to oxidation and is lighter than copper. The Radiation shield installed on the cooler is shown in Figure 7.

Feed Window Design

The vacuum window is a critical component in the package design. A foam backed membrane window was chosen for the prototype design. In this approach the film strength of a material such as Kapton provides the vacuum seal and the mechanical strength. Low-loss low-density foam acts as a thermal insulator and provides additional compressive strength.

To determine the required Kapton thickness Burst tests were performed on a typical clamped ring joint. The expected burst pressure for a 14 cm Kapton window was extrapolated from burst measurements of 2.2 cm and 9 cm. To provide a burst pressure safety factor of 3 the window would require 0.08 to 0.13 mm Kapton film.

Kapton windows lose mechanical strength when exposed to the sun. To solve this problem a thin Teflon cloth is used as an outer layer over the Kapton. Gore manufactures a reduced density Teflon cloth with an advertised 30-year life in radome service.

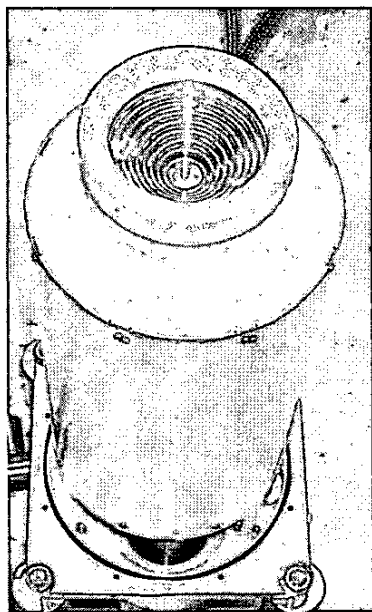


Figure 7 - Radiation Shield Installed

The prototype window is constructed of 0.08 mm Kapton covered with 0.38 mm Teflon cloth and is backed by 25 mm of polystyrene foam. The resulting noise temperature of the window is less than 1K. Figure 8 shows the cooler/LNA with the window installed.

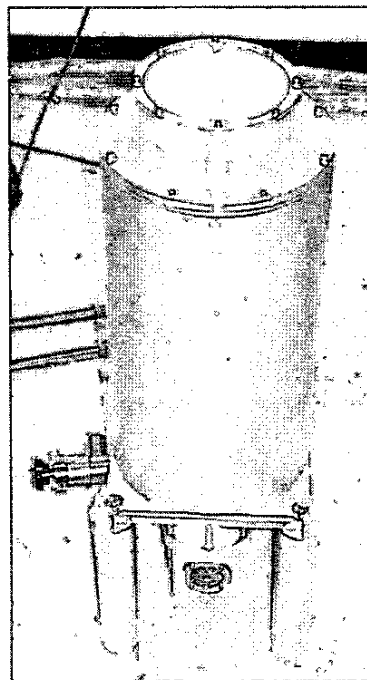


Figure 8 - Vacuum Window Installed on LNA Assembly

Vacuum System

The vacuum system is a critical component of the array LNA. A vacuum pressure of $<10^{-5}$ Torr is required in the vacuum space to minimize the heat leak to the cooled components. The array LNA uses a "cryopump" to maintain the vacuum pressure. A schematic of the vacuum system is shown in Figure 9.

Choice of Materials for Cooled Microwave Hardware—

Copper is normally used for most DSN microwave components. It provides the highest thermal conductivity (5w/cm-K@ 15K) of any practical material. It also has the lowest surface resistivity that affects the noise temperature contribution of microwave components. The noise temperature contribution to an LNA system associated with input waveguide loss is affected by two factors:

- 1) To the first-order the noise contribution of any given conductor is proportional to the absolute temperature from the equation: $T_{noise} = T_{physical} (1 - loss)$.
- 2) The loss is also proportional to the square root of the material surface resistivity that decreases with temperature.

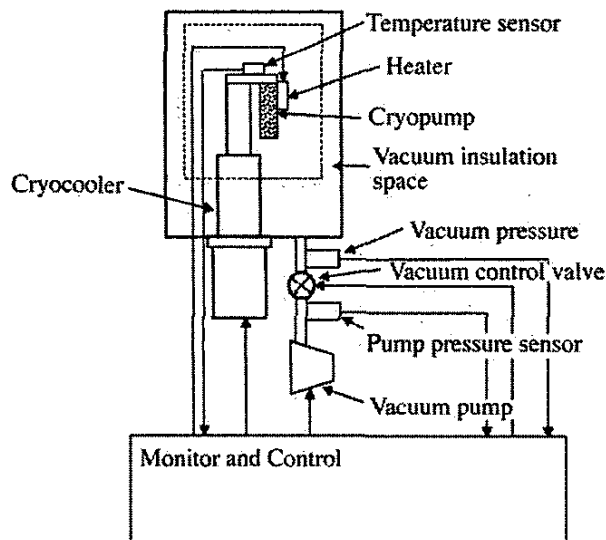


Figure 9 - Vacuum and Cryogenic System Schematic

The DSN typically specifies copper for waveguide in LNA systems. In the large array systems there may be an advantage in using aluminum waveguide to reduce manufacturing cost and the mass of feed components. The waveguide size is WR-112 for 8.4 GHz and WR-28 for 32 GHz. At the expected operating temperature of 20K or less the difference in loss of copper and aluminum waveguide is insignificant. Aluminum waveguide can therefore be used as desired.

5. MECHANICAL SUBSYSTEM:

The critical technology in the mechanical system for the reflector is the dish manufactured from a process called hydroforming. This is the process of forming aluminum to a rigid and precise mold by using a fluid or gas under pressure. It has been highly developed for use in production of low-cost reflectors for satellite communications and thousands of antennas in the 1 to 4 meter range have been manufactured. The advantages are: 1) High rigidity due to the one-piece aluminum shell (consider the stiffness of thin metal bowls or woks compared to the stiffness of flat sheets) 2) Accuracy largely determined by the mold rather than human error. The repeatability of the process was verified by fabricating 3 dishes with an rms of less than 0.2 mm and 3) Low costs for both raw material and labor.

For the DSN breadboard, the dish aperture is 6.048 m or 20 ft (see Figure 10). The dish is connected to a rigid Truss Structure at two places. The dish is hard mounted to the Truss Structure at its center. Spars connect the rim of the Dish to the rear of the Truss Structure. The Truss Structure is connected to the Petal Yoke at the elevation pivot point. There is a linear actuator or Jackscrew mechanism attached to the rear of the Yoke. As the Jackscrew extends or contracts the elevation of the Main Dish is changed. The Yoke is connected to the Petal Base through a Slew Bearing,

with gears on the outer ring. Two opposing motors, mounted inside the Yoke, drive the Azimuth Axis. The Dish, Truss Structure and Spars are made of aluminum. The Yoke and Pedestal Base are made of steel. The total weight of the antenna is approximately 8500 lbs. See Figure 11 for a drawing of the complete antenna.

Performance Requirements

Surface accuracy of the dish after manufacturing must be less than 0.2 mm rms. For either gravity, wind or temperature the change in rms cannot be greater than 0.13 mm. The antenna is required to operate year round at temperatures from -10 to 55C. The pointing accuracy must be less than 0.01 degrees. The rigid body displacement of the Main Dish between Antennas must be less than 0.5 mm. Displacement of the Subreflector relative to the Main Dish must be less than 0.5 mm. Displacement of the Feed relative to the Main Dish must be less than 1 mm axial and 0.5 mm radial. The antenna must have a range of 10 to 90 degrees in elevation and 0 to 450 degrees in Azimuth. The antenna is designed to operate up to 30 mph winds, can drive to stow in a 50 mph wind and survive a 100 mph wind.

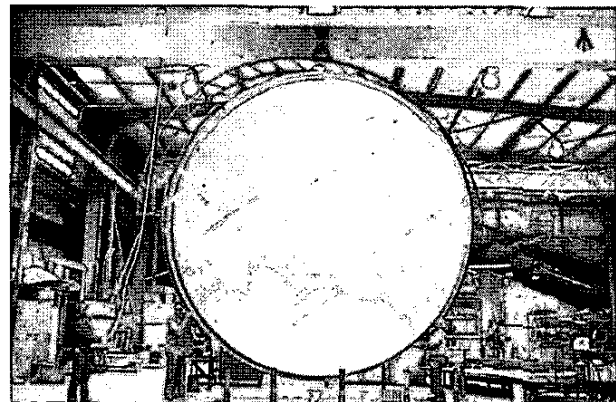


Figure 10 - Hydroformed Dish

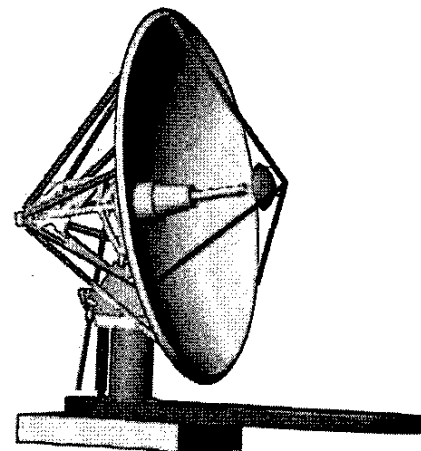


Figure 11 - The 6-meter Breadboard Antenna

Truss Structure & Spars

The Truss Structure is the main load-bearing substructure for the Antenna. It shares an interface with the Dish, Feed, Spars and Pedestal. The Elevation axis is located at the lower part of the Truss Structure. The loads from the jackscrew are reacted through the truss structure. The Spars, Truss Structure and Main Dish are a structural system. They work together to give the system rigidity. The Spars minimize deflection at the outer rim of the dish by transferring loads to the rear of the Truss Structure:

Feed Tower

The Feed Tower has two main structures. The cylindrical section houses the feed horn and waveguide. The conical section houses some of the electronics. The conical section is removable to allow access to service the Feed without disconnecting it from the Antenna. The cantilever loads from the feed are reacted into the Truss Structure. The Feed can be removed from the antenna without disconnecting the Dish from the Truss Structure.

Pedestal

The Pedestal consists of a Yoke and Base. The Base is stationary while the Yoke rotates about the long axis of the Base. There is a large bearing with gears that connect the Yoke to the Base. Motors mounted inside the Yoke drive the azimuth axis. There are pivots at the top of the Yoke about which the Dish and Spar rotate in the elevation axis.

Cable Management System

The azimuth axis range of motion requirement is 450 degrees. The Cable Management System forces the cable bundle to expand and contract in a planar spiral pattern. A rigid tube is connected to the Yoke. This Tube extends from the Yoke down to the bottom of the pedestal. The bottom of the tube is flared to act as a stress relief for the cable. The cable bundle is routed through this tube.

Analysis

Finite Element Analysis (FEA) was used to predict the performance of the antenna with gravity, wind and thermal loads. For any given load case the deformations of the Dish's reflective surface is extracted. Path length rms of these deflections is then determined. These predictions include the entire azimuth and elevation range. In addition, the frequencies of various bending modes were calculated. The FEA model is shown in Figure 12 and some typical results in Table 5. Figure 13 shows the two lowest order resonance modes.

Servo Motors and Actuators

Both the Azimuth and Elevation axis have the same Motors. The Motors are DC Servo with 110V AC input. The maximum horsepower is 2.0 hp. The stall torque is 165 in*lbs. These motors can run continuously at 50 in*lbs and 2000 rpm. The motors have 3000 counts per rev encoders built in. Also, each

Motor has a 240 in*lb brake. There are external encoders attached at each axis. These encoders have 144000 counts/rev and can resolve 9 arc seconds or .0025 degrees. On the azimuth axis there are two motors that work together to eliminate backlash. The motors maintain a constant counter torque. These two motors are connected to an 87:1 reducer. A pinion is connected to the output of the reducer. The motor reducer pinion combination drives a large Slew Bearing.

A linear actuator or Jackscrew is attached the rear of the yoke on a pivoting gimbal. The Jackscrews have a 20-ton capacity and require 48 revs at the input for 1.0 inch of raise. The previously described motor drives a geared belt and pulley system which in-turn drives the Jackscrew. The ratio of the geared belt and pulley system is 1.895:1. To reduce Jackscrew backlash, a 2000 lb ballast mass is attached to the rear of the Truss Structure. This ballast mass keeps the linear actuator in compression for all gravity and wind load cases. The Azimuth axis drive train and elevation axis jackscrew are shown in Figure 14.

6. COMPOSITES AS AN ALTERNATIVE ANTENNA DESIGN

With the increased popularity of composite materials, newer technologies and lower materials costs, composite materials are now an affordable, viable alternative for structural fabrication of small-to-medium size antennas.

Antennas can be fabricated with integrated backup structures to provide the stiffness required for high accuracy dish surfaces and reducing the fabrication process.

Since the thermal coefficient of expansion of carbon fiber composite materials is almost negligible, another great advantage for using this material is the significant reduction of thermal effects to avoid undesirable distortions of the reflective surface.

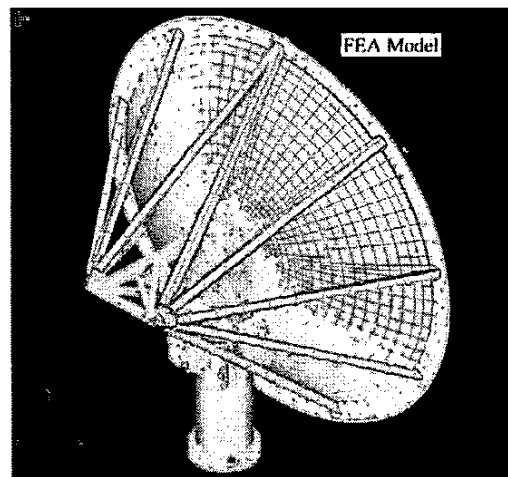
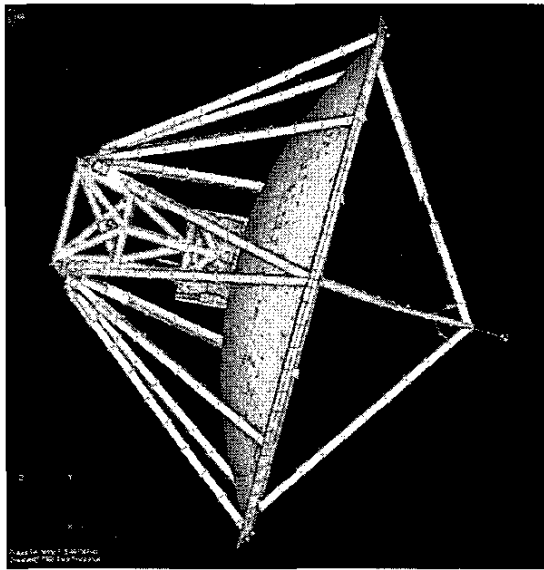


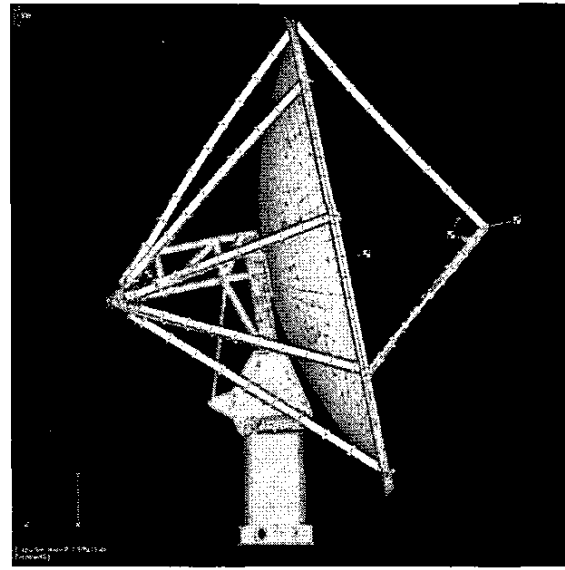
Figure 12 - The FEA Model

Table 5. Typical Results from the FEA Model

Gravity Load Elevation = 0 deg Azimuth = 0 deg					
x-Tilt, deg	y-Tilt, deg	Surface rms Including focus Error, mm	x-Translation, mm	y-Translation, mm	z-Translation, mm
-0.03	0	0.019	0	0	-0.55
Gravity Load Elevation = 60 deg Azimuth = 0 deg					
-0.04	0	0.024	0	0.74	-0.86

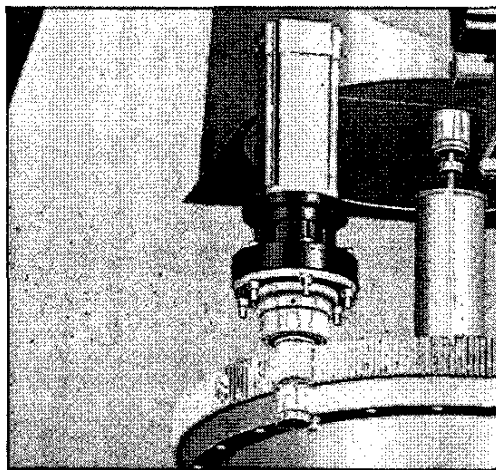


Mode 1. 5.46 Hz
(rotation about azimuth axis)

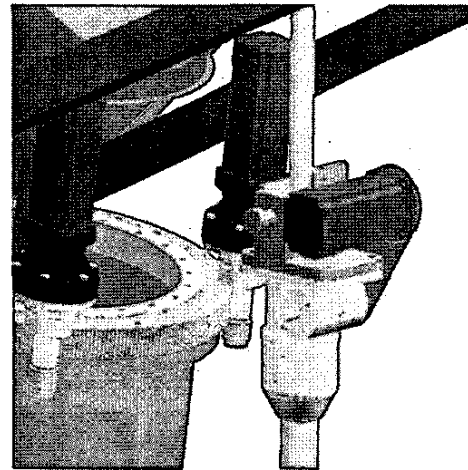


Mode 2. 7.08 Hz
(rocking for-and-aft)

Figure 13 - The Lowest Order Resonance Modes



Azimuth Axis
Drive Train



Elevation Axis Jackscrew
& Motor Pulley System

Figure 14 - Azimuth Axis Drive Train and Elevation Axis Jackscrew

Also, composite antennas provide a lighter structure with high rigidity, which helps in reducing the gravity deformations of the antenna.

The low cost solution for the array antennas is to use one of the various infusion technologies that have been recently developed by composite structures fabricators. Some of the advantages include low cost process (room temperature curing, no need for an autoclave) and materials (prepegs), and high material utilization factors; composite infusion is a repeatable, controlled process that uses low cost tooling. The infusion transfer processes include: Vacuum Assisted Resin Transfer Molding (VARTM), Resin Transfer Molding (RTM) and Resin Film Infusion (RFI).

To add stiffness to the structure without compromising the weight, the cross-section of the structure may be increased by using lightweight foam or metallic cores sandwiched between the carbon fiber laminates.

Several approaches are being investigated to provide the reflective structure of the dish; these include embedding conductor materials onto the carbon fiber laminates, aluminum foil and meshes integrated in the material.

7. SUMMARY

Currently, three antennas are under construction at JPL. The 3 dishes have been received, and the truss structure and spars have been fabricated. The feed tower and pedestal are under construction. A prototype feed was fabricated and tested. The breadboard feeds are under construction. A prototype cryogenic system has been fabricated and tested, with the breadboard systems under construction.

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A few years ago, he was supporting the Advanced Radar Technology Program working on inflatable radars and composite structures for advanced radar applications. He continues to support the Gossamer Technology program working on inflatable planar arrays for radar and communication antenna applications.

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Chris Porter — Biography not available.